



**ARTICLE**

# Numerical Simulation of the Wake Generated by a Helicopter Rotor in Icing Conditions

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## ABSTRACT

The wake generated by the rotor of a helicopter can exert a strong interference effect on the fuselage and the horizontal/vertical tail. The occurrence of icing on the rotor can obviously make this interplay more complex. In the present study, numerical simulation is used to analyze the rotor wake in icing conditions. In order to validate the overall mathematical/numerical method, the results are compared with similar data relating to other tests; then, different simulations are conducted considering helicopter forward flight velocities of 0, 10, 20, 50, and 80 knots and various conditions in terms of air temperature (atmospheric temperature degrading from  $-12^{\circ}\text{C}$  to  $-20^{\circ}\text{C}$  or from  $-20^{\circ}\text{C}$  to  $-26^{\circ}\text{C}$ ). The results indicate that the rotor aerodynamic performance (i.e., the lift-to-drag ratio distribution of the rotor disc) drops significantly once the rotor undergoes ice accretion. More importantly, the icing exerts a different influence of the wake dynamics depending on the atmospheric conditions. Interestingly, the rime-ice firstly occurs on the inner portion of rotor blades and then diffuses outward along the blade radial direction with the decrease in atmospheric temperature.

## KEYWORDS

Numerical simulation; helicopter; rotor wake; ice accretion

## Nomenclature

$a_0$	=	Rotor coning coefficient
$a_{1c}$	=	Rotor first-order longitudinal flapping angle
$a_{1s}$	=	Rotor first-order lateral flapping angle
$C_l, C_d$	=	Lift coefficient of rotor-blade airfoil and drag coefficient of the rotor-blade airfoil, respectively
$C_T, C_H$	=	Rotor thrust and horizontal force coefficients
$C_Y, C_Q$	=	Rotor side-force and torque coefficients
$\bar{e}, \sigma$	=	Blade flapping hinge offset and rotor solidity, respectively
$f$	=	Function symbol
$g$	=	Gravitational acceleration
$I_b, M_s$	=	Blade rotational inertia and blade mass, respectively
$m_G$	=	The mass of the helicopter
$N_r$	=	The number of segments along the direction of the blade-spanwise
$N_c$	=	The number of segments along the direction of the blade-chordwise



$N_w =$	The number of segments along the direction of the vortex age angle of the near wake sheet
$N_{fw} =$	The number of the vorticity straight-line segments
$N_\psi =$	The number of segments along the direction of the azimuthal angle of the rotor disc
$P_R, Q_R =$	The rotor roll and pitch angular velocity
$\bar{r}, R =$	The non-dimensional local radial position along rotor blade, and rotor radius, respectively
$T_{\text{Euler}} =$	The Euler transformation matrix
$\bar{u}_T, \bar{u}_P =$	The relative wind velocity from forward flight, and upward flapping at the blade section, respectively
$\bar{v}_i =$	The non-dimensional induced velocity of the rotor disk
$\beta, \beta_* =$	The rotor blade flapping angle and blade section inflow angle, respectively
$\rho, c =$	The air density and rotor blade chord, respectively
$\psi, \Delta\psi =$	The azimuthal angle of the rotor disc and azimuthal increment, respectively
$\xi =$	The vortex age angle of the vortex filament of each blade
$\Omega =$	The rotational speed of the rotor
$\Delta =$	The increments
$\theta_B =$	The pitching angle of the helicopter
$\phi_B =$	The rolling angle of the helicopter
$\psi_B =$	The yawing angle of the helicopter
$\tau, T_S =$	The icing time and atmospheric temperature, respectively
$LWC =$	The liquid water content
$MVD =$	The median volumetric diameter
$F, M =$	The force vector and force moment vector that acts on the helicopter
$H_B =$	The angular momentum vector of the helicopter
$r, V =$	The position and velocity vector of the vortex filaments
$u_\infty =$	The velocity vector of the free stream
$u_{fw} =$	The induced velocity vector of the far wake vortex filament
$u_{fw}^{bb} =$	The velocity vector of the far wake vortex filaments caused by the rotor blade motion
$V_B =$	The velocity vector of the helicopter
$\omega_B =$	The angular velocity vector of the helicopter

### ***Superscripts***

b =	The rotor blade-attached vortex
w, fw =	The near wake vortex lattices and far wake vortex filaments, respectively
' =	The ice accretion
T =	The matrix transposition

### ***Subscripts***

B =	The helicopter
FUS =	The fuselage of the helicopter
G =	The gravity of the helicopter
HT =	The horizontal tail of the helicopter
R, TR =	The main rotor and tail rotor of the helicopter, respectively
VT =	The vertical tail of the helicopter

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## 1 Introduction

Rotor blade ice accretion is a great hazardous factor that requires attention in helicopter design. This mainly affects the rotor aerodynamic performance and rotor wake feature, increases the complexity of aerodynamic interference between the rotor and fuselage, as well as the horizontal/vertical tail, and degrades the flight performance and flying qualities, threatening helicopter flight safety. Understanding the sensitivity of ice accretion and aerodynamic performance, as well as rotor wake feature and flight dynamic characteristics degradation, are essential in helicopter certification programs to ensure safety during flight in icing conditions. Early in 1974, a series of simulated and natural icing tests have been conducted to determine the capability of helicopters to operate in icing conditions [1]. Most researches have focused on the effects of icing on rotor aerodynamic characteristics [2–6], flight dynamic characteristics [7,8], and anti-icing system design [9–12], etc. Recently, a continuous in-depth study of the icing problem has been conducted [13–16]. Study on the behavior of the tip wake of a wind turbine that is similar to the rotor has been also conducted but lack of considering icing working conditions [17]. So far, the helicopter rotor wake feature that mainly affects the aerodynamic interference in icing conditions has been rarely conducted.

Without considering rotor icing, the free-vortex wake method is capable of directly modeling the helicopter rotor wake feature. This can be performed in various ways, such as by means of constant vorticity straight-line filaments [18], curved vortex filaments [19], or vortex blobs [20]. The straight-line segment approximation approach is most often used, because the induced velocity contribution of each segment can be evaluated exactly using the Biot-Savart law, and involves no approximate treatment. In terms of the numerical solution method, Bagai et al. [21] developed a classical pseudo-implicit predictor-corrector (PIPC) method for the solution of the rotor free-vortex wake problem early in 1995. Recently, the free-vortex wake method has been widely used. This can be combined with other analytical/numerical methods to explore more complex problems of helicopters, including rotor flowfield analysis [22], rotor aerodynamic interference [23], rotor aeroelastic analysis [24], etc. In addition, the rotor free-vortex wake method has also been incorporated in the helicopter flight dynamics model for high performance helicopter design [25–29].

In icing conditions, the free-vortex wake method should at least be combined with the rotor icing model, in order to simulate the helicopter rotor wake feature. Further considering the inflight helicopter trim problem embedded with the free-vortex wake method, more considerations might be developed, such as the rotor discrete aerodynamic model and complete helicopter flight dynamics model.

The present study presents a numerical simulation approach to simulate the helicopter rotor wake in icing conditions. First, the rotor free-vortex wake model was adopted to simulate the rotor wake geometry. Then, a numerical rotor discrete aerodynamic model was developed to solve the rotor force and moment acting on the helicopter. In addition, a rotor icing model was integrated into the discrete rotor aerodynamic model, which allows the model to predict the increments of iced rotor force, and torque and rotor flapping coefficients. In order to solve the iteration of the non-uniform distribution of rotor induced velocity and rotor flapping coefficients, a Crossed Coupling Iteration (CCI) algorithm was proposed. Furthermore, a helicopter flight dynamics model was also developed to conduct the inflight helicopter trim calculation. The rotor wake geometry of a helicopter at different flight velocity was numerically simulated and demonstrated, and the results and analysis were developed. Finally, a summary was presented.

## 2 Mathematical Model

### 2.1 Rotor Free-Vortex Wake Model

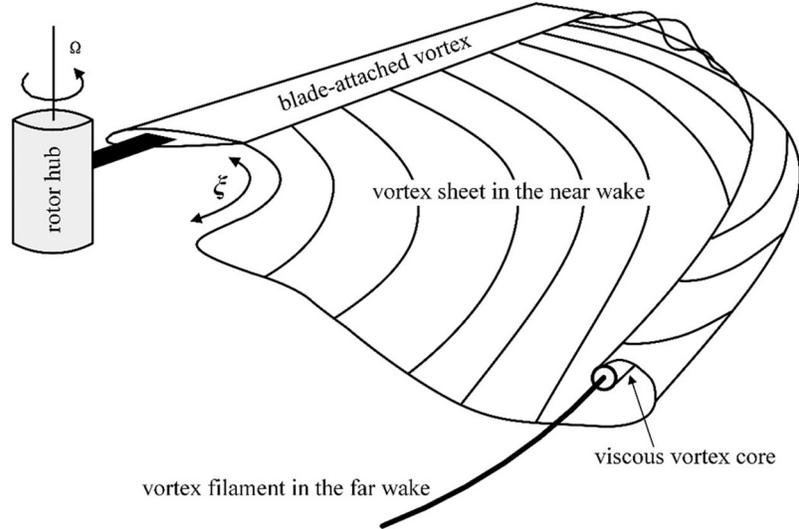
The rotor free-vortex wake model was developed to predict the rotor wake in icing conditions in the present study, based on the lifting surface theory and vortex method. The circulation of the blade-attached vortex was calculated using the lifting surface theory, and the lifting surface was arranged in the blade

mid-arc surface. Along the blade-span wise direction, the lifting surface was divided into  $N_r$  parts. In the blade-chordwise direction, this was divided into  $N_c$  parts. Then, the entire lifting surface of each rotor blade was divided into  $N_r \times N_c$  vortex lattice.

According to the rotor vortex wake method, the rotor wake can be classified as near wake and far wake. Near wake is fixed in the tangent plane, which is located at the trailing edge of the blade mid-arc surface. In this region (Fig. 1), the tip vortex is in the process of rolling up, while the inner vortex sheet is modeled in the form of quadrilateral lattice. The vortex sheet can be divided into  $N_r \times N_w$  vortex lattice, with  $N_r$  stations along blade-span wise direction for each of  $N_w$  azimuthal sectors. According to the Kutta Condition [30], the circulation of each vortex lattice on the blade trailing edge at the azimuth of  $\psi$  is equal to the value of the corresponding vortex lattice circulation on the blade trailing edge at the azimuth of  $\psi - \Delta\psi$ . In the far wake region, the blade tip vortices roll up, forming a dominant vortex filament with a high swirl velocity and a relatively small viscous vortex core (Fig. 1). The circulation can be taken as the peak amount of blade-attached vortex. The vortex filament of each blade can be divided into  $N_{fw}$  vorticity straight-line segments. These would move freely as material lines, according to Helmholtz's second law. The governing equation can be written, as follows:

$$\frac{\partial r}{\partial \psi} + \frac{\partial r}{\partial \xi} = \frac{1}{\Omega} \mathbf{V} = \frac{1}{\Omega} (\mathbf{u}_{\infty} + \mathbf{u}_{fw}^{bb} + \mathbf{u}_{fw}^b + \mathbf{u}_{fw}^w + \mathbf{u}_{fw}^{fw}) \quad (1)$$

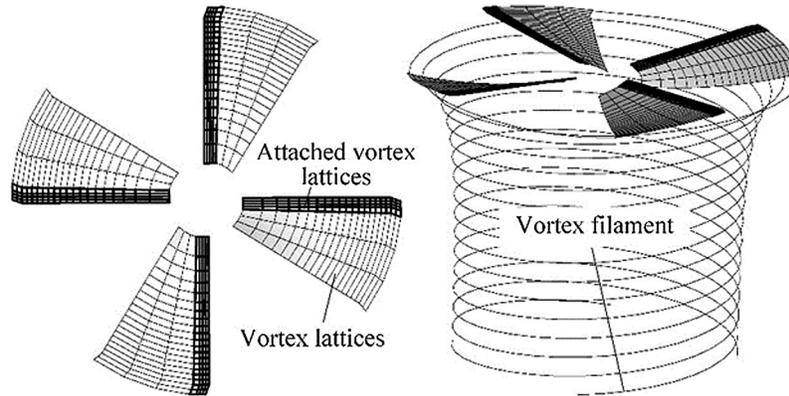
In Formula (1), the  $\mathbf{u}_{\infty}$  and  $\mathbf{u}_{fw}^{bb}$  are correlated to the helicopter flight status, including the rotor blade motion. These can be determined through the trim calculation of the following helicopter flight dynamics model. Fig. 2 shows a sample of the convergent rotor wake of the UH-60A helicopter in hover.



**Figure 1:** Sketch of the rotor wake formation

## 2.2 Rotor Icing Model

Considering the rotor icing, an engineering rotor icing model needs to be developed. This mainly involves the introduction of the icing-related increments of rotor thrust, horizontal force, side force, torque coefficients, and icing-related increment of flapping coefficients into the uniced flight dynamics model.



**Figure 2:** Sketch of the rotor wake description ( $N_r = 20$ ,  $N_c = 4$ ,  $N_w = 3$ ,  $N_{fw} = 144$ )

The basic lift and drag coefficients of the rotor-blade airfoil due to icing are, as follows:

$$\begin{cases} C'_l = (1 + \Delta C_l / C_l) C_l \\ C'_d = (1 + \Delta C_d / C_d) C_d \end{cases} \quad (2)$$

The coefficients for rotor thrust, side force, horizontal force, and torque due to icing are, as follows:

$$\begin{cases} C'_T = C_T + \Delta C_T \\ C'_Y = C_Y + \Delta C_Y \\ C'_H = C_H + \Delta C_H \\ C'_Q = C_Q + \Delta C_Q \end{cases} \quad (3)$$

The iced-rotor flapping model in the Fourier series form can be presented, as follows:

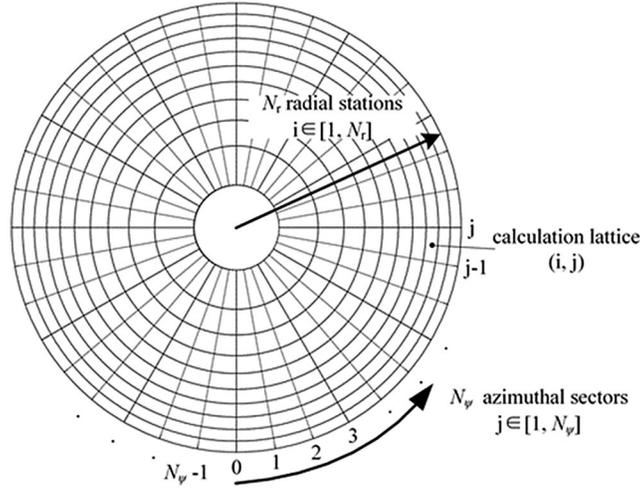
$$\begin{cases} a'_0 = a_0 + \Delta a_0 \\ a'_{1c} = a_{1c} + \Delta a_{1c} \\ a'_{1s} = a_{1s} + \Delta a_{1s} \end{cases} \quad (4)$$

In Formulas (2), (3), and (4), the rotor thrust, horizontal force, side force and torque coefficients, as well as the rotor flapping coefficients, were calculated using the following Rotor Discrete Aerodynamic Model. Furthermore, the corresponding increments caused by icing can be calculated by the established engineering icing model in [7] and [8].

### 2.3 Rotor Discrete Aerodynamics Model

The rotor discrete aerodynamic model was developed to embed the rotor free-vortex wake model and rotor icing model into the following helicopter flight dynamics model. In the process of modeling and numerical simulation, based on the principle of equal annular area, the rotor disc was divided into  $N_r \times N_\psi$  calculating sectors, with  $N_r$  radial stations for each of the  $N_\psi$  azimuthal sectors, as shown in Fig. 3.

The discrete expressions of the rotor thrust, horizontal force, side-force, and torque coefficients can be deduced, as follows:



**Figure 3:** Sketch of the rotor disc discretization ( $N_r = 10$ ,  $N_p = 36$ )

$$\begin{pmatrix} C_T \\ C_H \\ C_Y \\ C_Q \end{pmatrix} = \frac{\sigma \Delta \psi}{4\pi} \sum_{i=1}^{N_r} \sum_{j=1}^{N_p} \begin{pmatrix} F_c^{(i,j)} \cos \beta^{(j)} \\ F_s^{(i,j)} \sin \psi^{(j)} - F_c^{(i,j)} \sin \beta^{(j)} \cos \psi^{(j)} \\ -F_s^{(i,j)} \cos \psi^{(j)} - F_c^{(i,j)} \sin \beta^{(j)} \sin \psi^{(j)} \\ -F_s^{(i,j)} (\bar{r}^{(i)} \cos \beta^{(j)} + \bar{e}) \end{pmatrix} \quad (5)$$

where,

$$F_c^{(i,j)} = \left( C_l^{(i,j)} \cos \beta_*^{(i,j)} - C_d^{(i,j)} \sin \beta_*^{(i,j)} \right) F_0^{(i,j)} \quad (6)$$

$$F_s^{(i,j)} = \left( C_l^{(i,j)} \sin \beta_*^{(i,j)} + C_d^{(i,j)} \cos \beta_*^{(i,j)} \right) F_0^{(i,j)} \quad (7)$$

$$F_0^{(i,j)} = \left[ \left( \bar{u}_T^{(i,j)} \right)^2 + \left( \bar{u}_P^{(i,j)} \right)^2 \right] \Delta \bar{r}^{(i)} \quad (8)$$

The  $C_l^{(i,j)}$  and  $C_d^{(i,j)}$  in icing conditions can be derived from the above rotor icing model, that is, by using Formula (2).

The equations for the rotor coning and first-order longitudinal and lateral flapping coefficients in the rotor flapping model are, as follows:

$$\begin{cases} f_1(a_0, a_{1c}, a_{1s}) = I_b \Omega^2 a_0 - M_{T0} + M_{sg} = 0 \\ f_2(a_0, a_{1c}, a_{1s}) = M_{Tc} + 2P_R \Omega I_b = 0 \\ f_3(a_0, a_{1c}, a_{1s}) = M_{Ts} - 2Q_R \Omega I_b = 0 \end{cases} \quad (9)$$

The expressions of  $M_{T0}$ ,  $M_{Tc}$  and  $M_{Ts}$  are, as follows:

$$M_{T0} = \frac{\Delta \psi}{4\pi} \rho c (\Omega R)^2 R^2 \sum_{i=1}^{N_r} \sum_{j=1}^{N_p} F_c^{(i,j)} \bar{r}^{(i)} \quad (10)$$

$$M_{Tc} = \frac{\Delta\psi}{2\pi} \rho c (\Omega R)^2 R^2 \sum_{i=1}^{N_r} \sum_{j=1}^{N_\psi} F_c^{(i,j)} \bar{r}^{(i)} \cos \psi^{(j)} \quad (11)$$

$$M_{Ts} = \frac{\Delta\psi}{2\pi} \rho c (\Omega R)^2 R^2 \sum_{i=1}^{N_r} \sum_{j=1}^{N_\psi} F_c^{(i,j)} \bar{r}^{(i)} \sin \psi^{(j)} \quad (12)$$

When Formula (5) was solved, the rotor coning and first-order longitudinal and lateral flapping coefficients can be obtained by solving the nonlinear equations, that is, Formula (9).

### 2.4 Helicopter Flight Dynamics Model

In the present study, a nonlinear flight dynamic model of a single rotor helicopter with a tail rotor was developed, as follows:

$$\begin{cases} m_G \dot{V}_B + m_G \omega_B \times V_B = F_B \\ \dot{H}_B + \omega_B \times H_B = M_B \\ \omega_B = T_{\text{Euler}} [\dot{\theta}_B, \dot{\phi}_B, \dot{\psi}_B]^T \end{cases} \quad (13)$$

In Formula (13), the resultant force vector ( $F_B$ ) and resultant moment vector ( $M_B$ ) that acts on the helicopter can be presented, as follows:

$$\begin{cases} F_B = F_R + F_{TR} + F_{FUS} + F_{HT} + F_{VT} + F_G \\ M_B = M_R + M_{TR} + M_{FUS} + M_{HT} + M_{VT} \end{cases} \quad (14)$$

In the above formula, the rotor force ( $F_R$ ) and moment ( $M_R$ ) vectors that act on the helicopter are the functions of rotor thrust, horizontal force, side-force and torque coefficients in the above rotor discrete aerodynamic model.

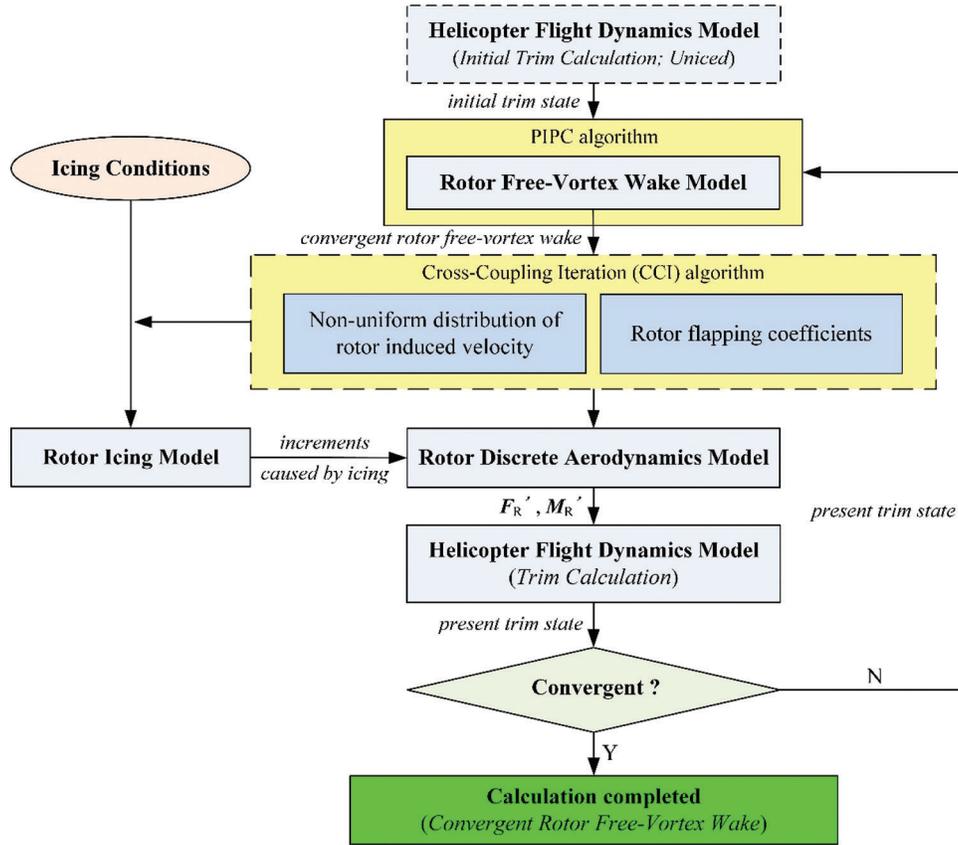
$$\begin{cases} F_R = f_1(C_T, C_H, C_Y, C_Q) \\ M_R = f_2(C_T, C_H, C_Y, C_Q) \end{cases} \quad (15)$$

In Formula (14), the tail rotor force ( $F_{TR}$ ) and force moment ( $M_{TR}$ ) vectors can be calculated using the traditional Rotor Momentum Theory [31]. The force and force moment vectors of fuselage ( $F_{FUS}$ ,  $M_{FUS}$ ), horizontal tail ( $F_{HT}$ ,  $M_{HT}$ ), and vertical tail ( $F_{VT}$ ,  $M_{VT}$ ) can be obtained using the wind tunnel test data [32].

### 3 Implementation Procedure

In the present study, the rotor free-vortex wake model and rotor icing model were incorporated into the rotor discrete aerodynamics model to output the  $F'_R$  and  $M'_R$  needed in the flight dynamics model. By trimming the flight dynamics model, the numerical simulation of the helicopter rotor free-vortex wake in icing conditions could be conducted.

As shown in Fig. 4, the initial uniced helicopter flight dynamics trim calculation can be conducted using the Rotor Vortex Model (RVM) [33]. Then, the rotor free-vortex wake iteration can be conducted to solve the convergent wake geometry. In addition, the non-uniform distribution of the rotor-induced velocity and rotor flapping coefficients can be obtained using the proposed Cross-Coupling-Iteration (CCI) algorithm. Subsequently, the rotor icing model predicts the increments of the iced rotor force and torque, as well as the rotor flapping coefficients caused by icing. Then, the calculation of the rotor discrete aerodynamics model can be conducted. When the trim calculation of the helicopter flight dynamics model becomes convergent, the numerical simulation of the helicopter rotor wake in icing conditions is completed. Otherwise, the rotor free-vortex wake iteration should be conducted again based on the present trim state.



**Figure 4:** Flowchart for the numerical simulation

## 4 Numerical Method

### 4.1 PIPC Algorithm

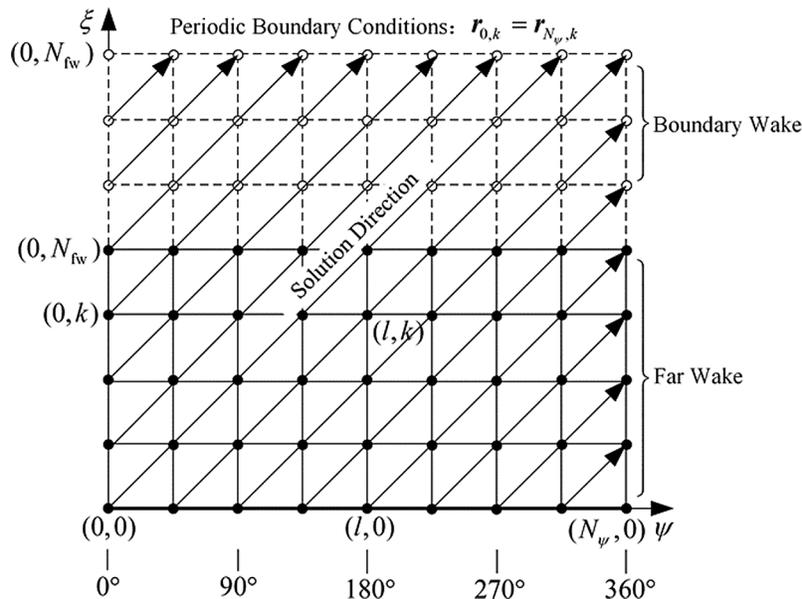
Bagai and Leishman's PIPC algorithm [21] was adopted to solve the above governing equation in Formula (1). Fig. 5 presents the solution domain, solution direction, grid setting, periodic boundary conditions, etc. For the numerical solution of the far wake in finite space, the far wake can be truncated using the concept of the boundary wake [34].

Using the five-point central difference method, the first order differential terms of the governing equation in Formula (1) can be presented, as follows:

$$\begin{cases} \left. \frac{\partial r}{\partial \psi} \right|_{(l+1/2, k+1/2)} = \frac{1}{2} \left( \frac{r_{l, k+1} - r_{l, k}}{\Delta \psi} + \frac{r_{l+1, k+1} - r_{l+1, k}}{\Delta \psi} \right) \\ \left. \frac{\partial r}{\partial \xi} \right|_{(l+1/2, k+1/2)} = \frac{1}{2} \left( \frac{r_{l+1, k} - r_{l, k}}{\Delta \xi} + \frac{r_{l+1, k+1} - r_{l, k+1}}{\Delta \xi} \right) \end{cases} \quad (16)$$

The  $\Delta \xi$  is equal to  $\Delta \psi$  for the convenience of the numerical solution. The velocity term ( $V$ ) of the governing equation in Formula (1) can be deduced as the mean velocity.

$$V(r_{l+1/2, k+1/2}) = [V(r_{l, k}) + V(r_{l, k+1}) + V(r_{l+1, k+1}) + V(r_{l+1, k})]/4 \quad (17)$$



**Figure 5:** Sketch of the solution domain and solution direction

Then, the governing equation in Formula (1) can be presented, as follows:

$$\mathbf{r}_{l+1, k+1} = \mathbf{r}_{l, k} + \frac{\Delta\psi}{4\Omega} [\mathbf{V}(\mathbf{r}_{l, k}) + \mathbf{V}(\mathbf{r}_{l, k+1}) + \mathbf{V}(\mathbf{r}_{l+1, k+1}) + \mathbf{V}(\mathbf{r}_{l+1, k})] \quad (18)$$

Since both the five-point central difference and mean velocity approximation have second-order accuracy, the above Formula (18) also has second-order accuracy for the governing equation in Formula (1). Based on the Formula (18), the governing equation can be solved through the calculation of the prediction and correction in each iteration using Bagai and Leishman's PIPC algorithm.

#### 4.2 Cross Coupling Iteration (CCI) Algorithm

In the present study, the Cross-Coupling-Iteration (CCI) algorithm was proposed to calculate the non-uniform distribution of rotor-induced velocity and the corresponding flapping coefficients (Fig. 6) for avoiding computational divergence.

In the CCI algorithm, two constraints were performed: the constraint on rotor-induced velocity iteration, and the constraint on rotor flapping coefficients iteration. When the non-uniform distribution of rotor-induced velocity iteration approaches to the divergence, the uniform distribution of rotor-induced velocity, based on the Rotor Momentum Model (RMM) [33], can be used to as the iteration values at the moment, in order to enter the next iteration. If the rotor flapping coefficients iteration is not convergent, the rotor flapping coefficients, based on the RMM, can be used to as the iteration values at the moment, in order to enter the next iteration.

#### 4.3 Verification of the Numerical Calculation Method

The validity and accuracy of the numerical calculation method was verified through comparison with the available experimental measurements [34]. The model rotor geometry and operating conditions for the rotor configuration is summarized in Tab. 1.

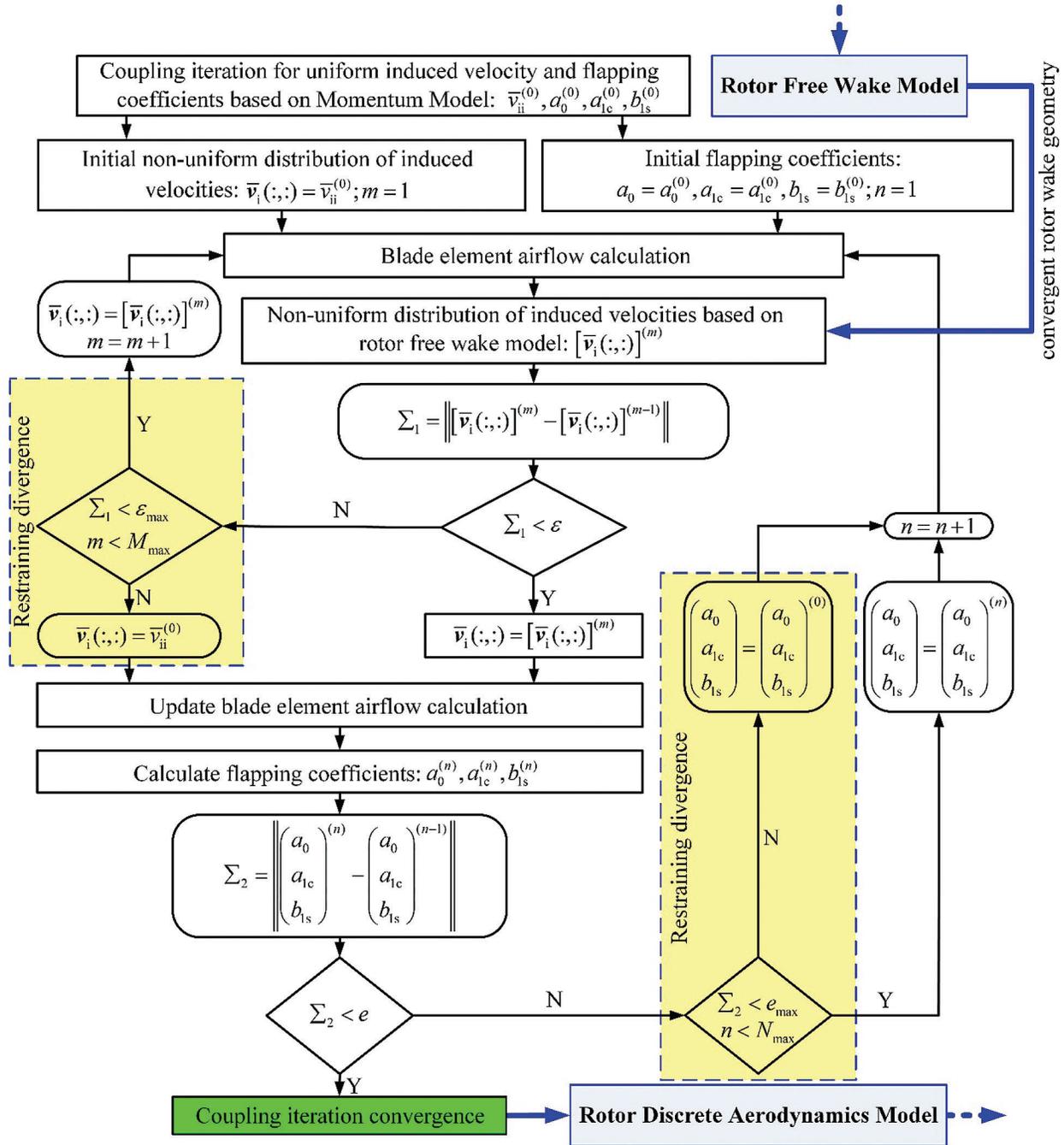


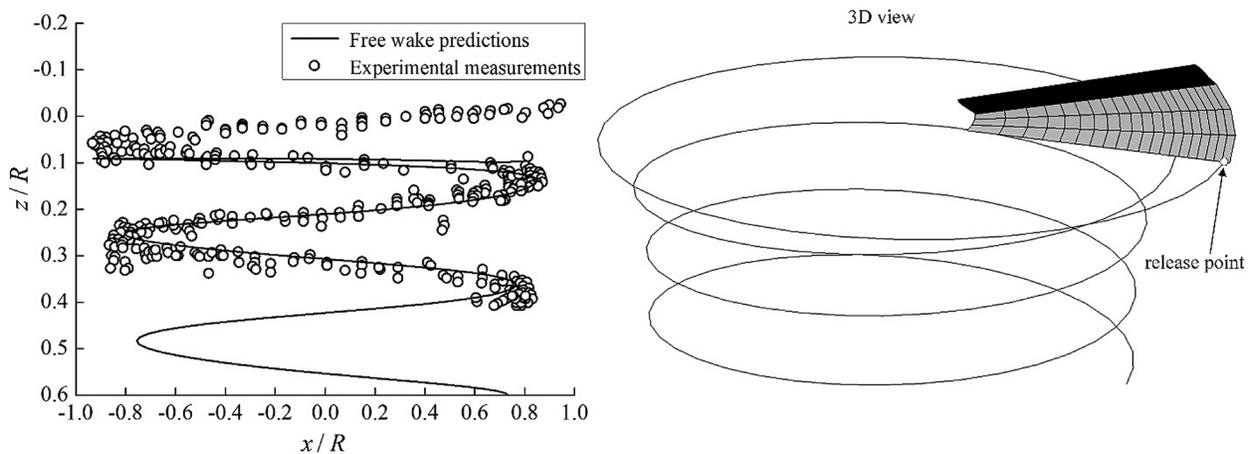
Figure 6: Flow chart for the CCI method

Fig. 7 presents the free-vortex wake predictions along with the experimental measurements. This indicates that the predictions and experimental measurements are in good agreement, except for some discrepancies in the wake age, which ranged from 0 to 180° (i.e., the first circle of the rotor wake). The reason for these discrepancies is that the release point (Fig. 7) of the blade tip vortex in the model is located on the quadrilateral vortex lattice of the near wake, while the near wake is fixed in the tangent plane of the trailing edge of blade mid-arc surface. Hence, the vortex sheet of the near wake occupies a fixed space along with the vertical down direction, allowing the rotor free-vortex wake of the first circle

to have a larger change, when compared to the experimental measurements. However, according to the results after the first circle of the rotor wake, the near wake setting method (i.e., fixed in the tangent plane of the trailing edge of the blade mid-arc surface) does not affect the calculation accuracy of the subsequent rotor free-vortex wake.

**Table 1:** Operating conditions for the rotor configuration

Blade number	1
Radius, m	0.4064
Chord, m	0.0425
Hinge offset, $\bar{e}$	0.0
Spar length, $\bar{r}_0$	0.2
Tip speed, m/sec	89.37
Airfoil section	NACA2415
Collective pitch, degree	4.0
Twist, degree	0.0
Thrust coefficient	0.003

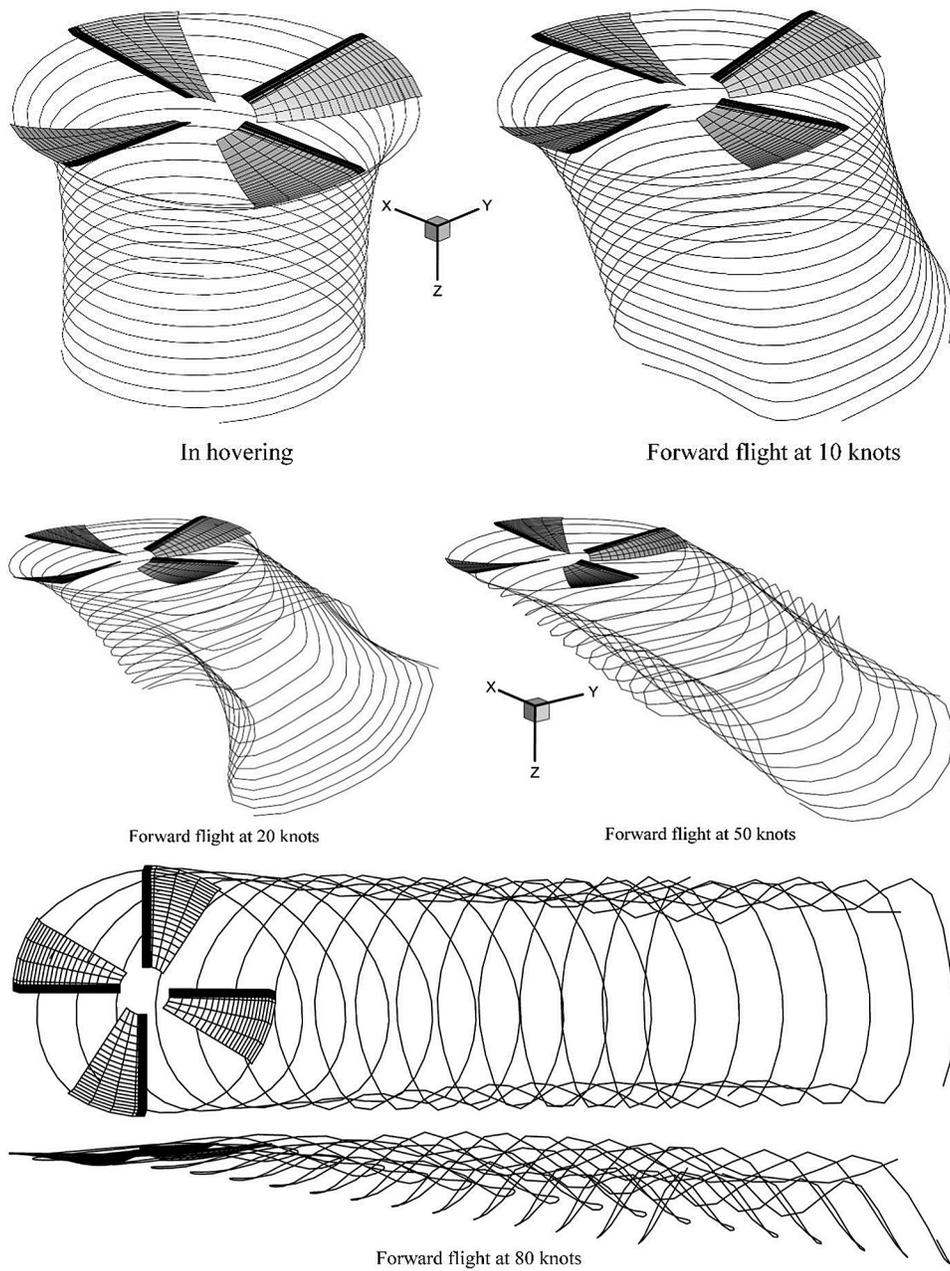


**Figure 7:** Free-vortex wake predictions along with the experimental measurements

Fig. 8 shows the convergent rotor wake geometries of the UH-60A helicopter in hovering, and at different forward flight velocities in clean ice conditions. This shows that the free-vortex wake method embedded in the helicopter flight dynamics model can well-capture the rotor wake geometric characteristics. Especially, with the increase of forward flight velocity, the process of the tip vortex rolling up of the rotor blade is well-demonstrated.

## 5 Results and Analysis

The effects of icing on the rotor-vortex wake due to icing time and atmospheric temperature were mainly analyzed using the above mathematical model and the corresponding procedure, as well as the numerical method. Merely the wake of one blade at the azimuth of zero degree was depicted and analyzed in detail for convenience. The basic icing conditions are presented in Tab. 2.



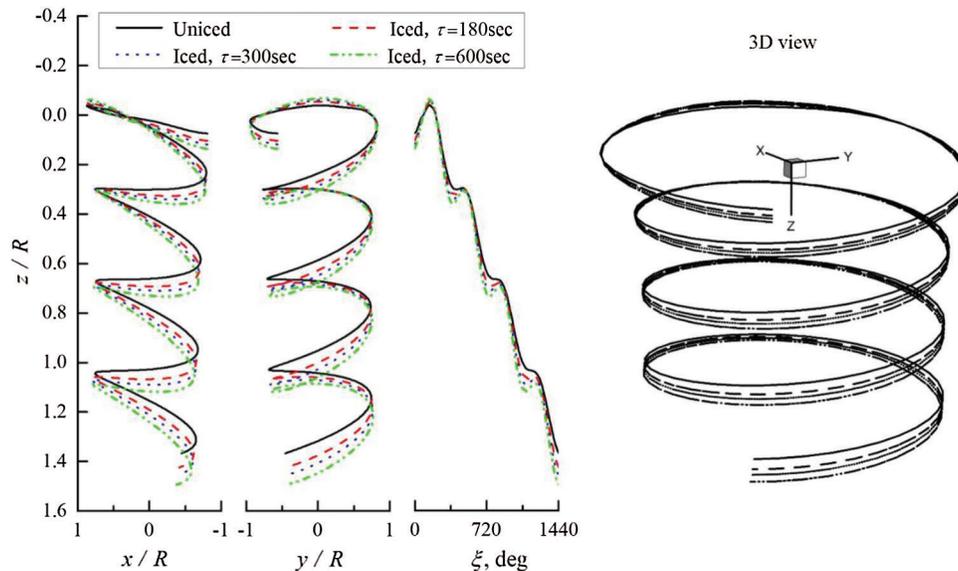
**Figure 8:** The rotor free-vortex wake geometries of the UH-60A helicopter at different flight velocities

**Table 2:** Basic icing conditions

$T_s, ^\circ C$	$LWC, g/m^3$	$MVD, \mu m$	$\tau, sec$
-26.0	1.0	20.0	180.0

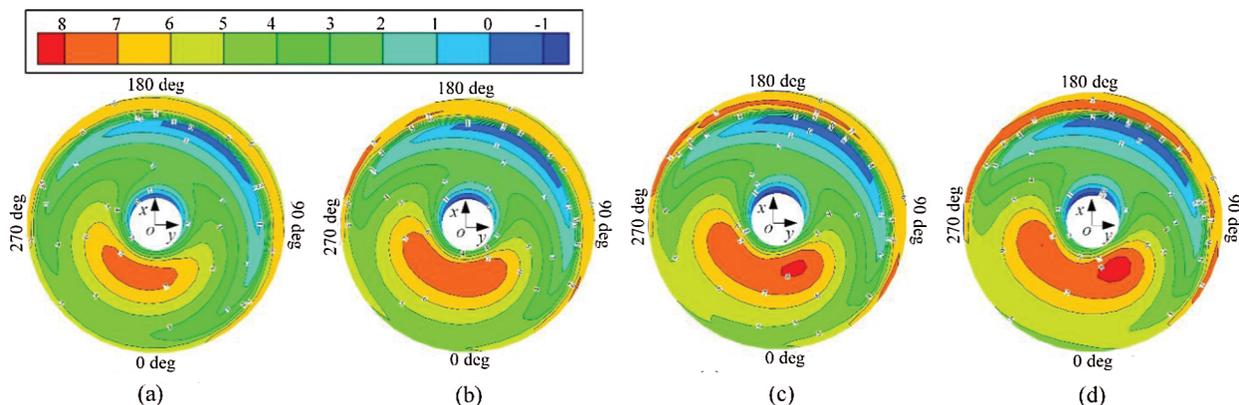
### 5.1 Icing Time

Fig. 9 presents the effects of icing on the rotor free-vortex wake due to icing time in hovering. In Fig. 9, the rotor icing makes the wake geometry slightly sink. This also induces the wake geometry to slightly move along the azimuth of 180 and 90 degrees. Furthermore, as the icing time increases, the variation trend become increasingly obvious. Furthermore, Fig. 9 presents the heights of the blade tip vortex filament versus vortex age angle, and the specific changes in the sunk amplitude of the free-vortex wake.



**Figure 9:** Effects of icing on the rotor free-vortex wake due to  $\tau$  in hovering

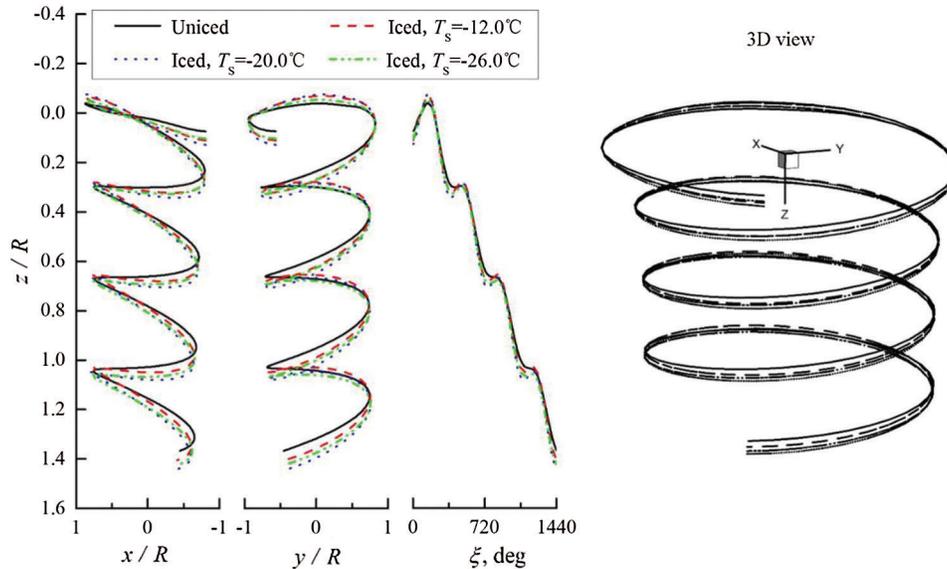
Some of the reasons for the rotor free-vortex wake changes with icing time might be due to the effects of icing on the angle of attack (AOA) of the rotor blade. In comparing the uniced distribution of AOA, Fig. 10 presents the effects of icing on the distribution of AOA due to the icing time, which ranged from 180 s to 300 s, and to 360 s. This indicates that the AOA of the rotor blade increases with the icing time. Furthermore, at the blade tip area in the front half part, and at the blade root area in the back half part of the rotor disc, the AOA of the rotor blade exhibits an evident increase with icing time. The reason for the icing effects on the AOA of the rotor blade might be due to the rotor performance degradation. In this situation, the inflight helicopter needs to maintain a stable hover by at least increasing the collective pitch control, in order for the AOA to increases.



**Figure 10:** Effects of icing on the AOA of the rotor blade due to  $\tau$  in hovering. (a) Uniced, (b) Iced,  $\tau = 180$  s, (c) Iced,  $\tau = 300$  s, (d), Iced,  $\tau = 360$  s

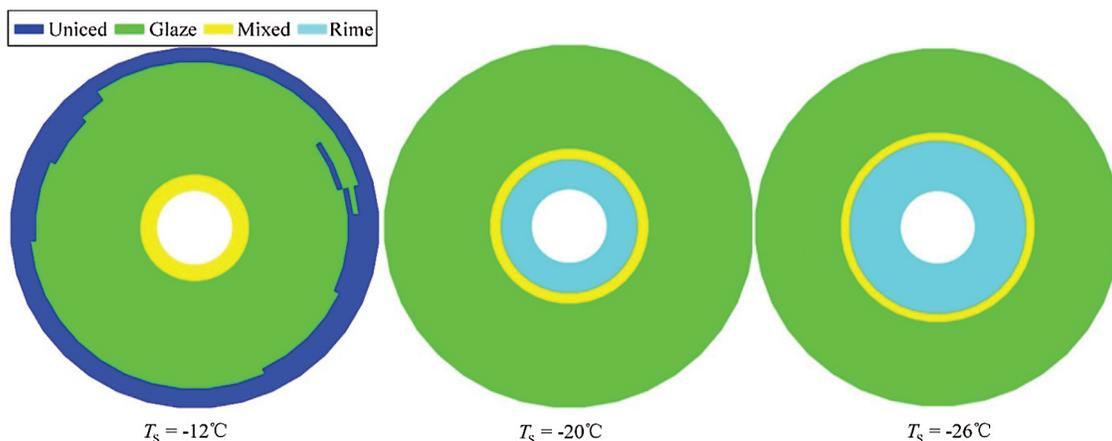
## 5.2 Atmospheric Temperature

Fig. 11 presents the effects of icing on the rotor free-vortex wake due to atmospheric temperature in hovering. As shown in Fig. 11, the rotor icing induced the free-vortex wake to slightly sink when the atmospheric temperature degraded from  $-12^{\circ}\text{C}$  to  $-20^{\circ}\text{C}$ . However, this caused the sunk trend to slightly recover when the atmospheric temperature continued to degrade from  $-20^{\circ}\text{C}$  to  $-26^{\circ}\text{C}$ . Fig. 11 presents the heights of the blade tip vortex filament versus vortex age angle, and the specific changes in amplitude of this trend.



**Figure 11:** Effects of icing on rotor wake due to  $T_s$  in hovering

Some of the reasons for this change trend might be due to the temperature effects. In general, the performance penalties caused by glaze ice is more serious than mixed ice, and subsequently by rime ice. The rotor icing tunnel test data indicated that as the temperature increased, the accreted ice shape on the outer portion of the rotor blades changed from rime to mixed, and subsequently to glaze, increasing the performance penalties [35]. In this section, with the use of the ice shape definition method [8], the distribution of the ice shape on the rotor disc at different atmospheric temperatures was calculated to validate the change trend of the effects of icing on the rotor free-vortex wake due to atmospheric temperature, as shown in Fig. 12.

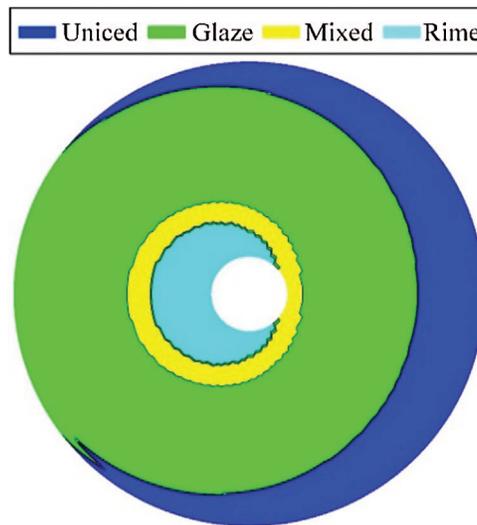


**Figure 12:** The distribution of the ice shape on the blades of the rotor disc in hovering

Fig. 12 shows that the range of the glaze ice that accreted on the outer portion of the rotor blades significantly increased when the atmospheric temperature degraded from  $-12^{\circ}\text{C}$  to  $-20^{\circ}\text{C}$ , increasing the performance penalties, and subsequently making the rotor downstream wake sink (Fig. 11). However, the range of the rime ice that accreted on the inner portion of the rotor blades evidently increased when the atmospheric temperature degrades from  $-20^{\circ}\text{C}$  to  $-26^{\circ}\text{C}$ , decreasing the performance penalties and making the rotor downstream free-vortex wake have a slightly reverted change trend to the uniced state (Fig. 11). The reason for this is that the performance penalties due to rime ice is lower than the glaze ice, and that this has an irregular shape. In addition, the uniced area on the blade tip region in Fig. 12 was generated by an extremely high local temperature, preventing the occurrence of ice accretion. The reason for this is that there might be a positive correlation between the high local temperature and the high local Mach number at the blade tip region.

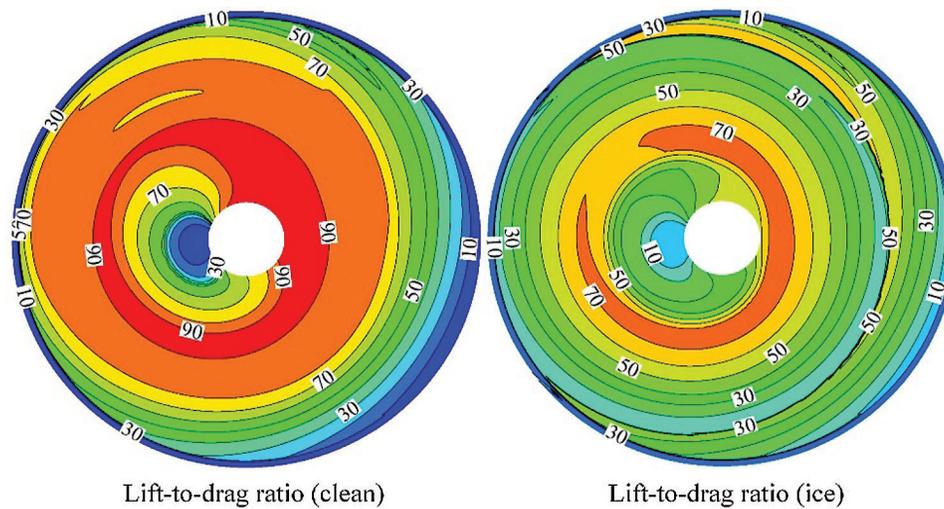
### 5.3 Forward Flight Velocity

For further comparison with the situation of the high local temperature at the blade tip region at hover in Fig. 11, Fig. 13 presents the distribution of the ice shape on the blades of the rotor disc at a forward flight of 60 knots, with  $T_s = -12^{\circ}\text{C}$ ,  $\tau = 60\text{ s}$  and  $LWC = 0.75\text{ g}/m^3$ . Compared with Fig. 12, when a helicopter changes from hover to forward flight, the icing area might gradually diffuses to the blade tip at the retreating blade region (left side direction of Figs. 12 and 13) due to the local temperature reduction on the retreating blade region, and the reduction in the local Mach number of the blade tip region at forward flight velocity.



**Figure 13:** The distribution of the ice shape on the blades of the rotor disc at a forward flight of 60 knots

Fig. 14 presents the comparison of the lift-to-drag ratio distribution of the rotor disc before and after icing at a forward flight of 60 knots. This evidently indicates that the lift-to-drag ratio evidently drops when the helicopter rotor encounters ice accretion at forward flight velocity.



**Figure 14:** The lift-to-drag ratio distribution of the rotor disc at a forward flight of 60 knots

## 6 Summary

By incorporating the rotor free-vortex wake model and rotor icing model into the rotor discrete aerodynamics model, and using the proposed Cross Coupling Iteration (CCI) algorithm, the trim calculation of the helicopter flight dynamics model in icing conditions was conducted, in order to acquire the convergent rotor free-vortex wake geometry, and investigate the effects of icing on the rotor free-vortex wake. The following conclusions that might have some significance in helicopter flight safety encounters with ice accretion can be mainly summarized, as follows:

1. The proposed numerical simulation method appears to be an adequate tool for helicopter rotor free-vortex wake analysis in icing conditions.
2. Both the sunk trend of the rotor free-vortex wake due to icing and the variation of the ice shape on the rotor blades might lead to rotor aerodynamic degradation: on one hand, ice accretion causes the rotor free-vortex wake to sink, and this sinking trend becomes more evident with the increase in icing time. On the other hand, a rime-ice shape initially occurs on the inner portion of rotor blades, and diffuses outward along the blade with the decrease in atmospheric temperature. In addition, the uniced area of the rotor blade tip is generated by the extremely high local temperature, preventing the occurrence of ice accretion. However, a glaze-ice shape occurs when the atmospheric temperature continues to decrease.
3. As shown in the forward flight velocity analysis, the lift-to-drag ratio would evidently drop when helicopter rotor encounters ice accretion. When the helicopter changes from hover to forward flight in the typical condition of  $T_S = -12^\circ\text{C}$ , the glaze ice might gradually diffuses to the blade tip on the retreating blade region, which is due to the decrease in local temperature on the retreating blade region, and the reduction in local Mach number at the blade tip region at forward flight. This phenomenon might provide a great enlightenment for the design of anti-icing and deicing of helicopter rotor blades.

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